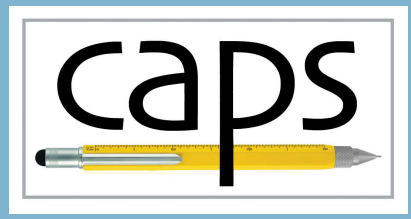
Computational Aircraft Prototype Syntheses



Training Session 3.5
CAPS Analysis

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Overview

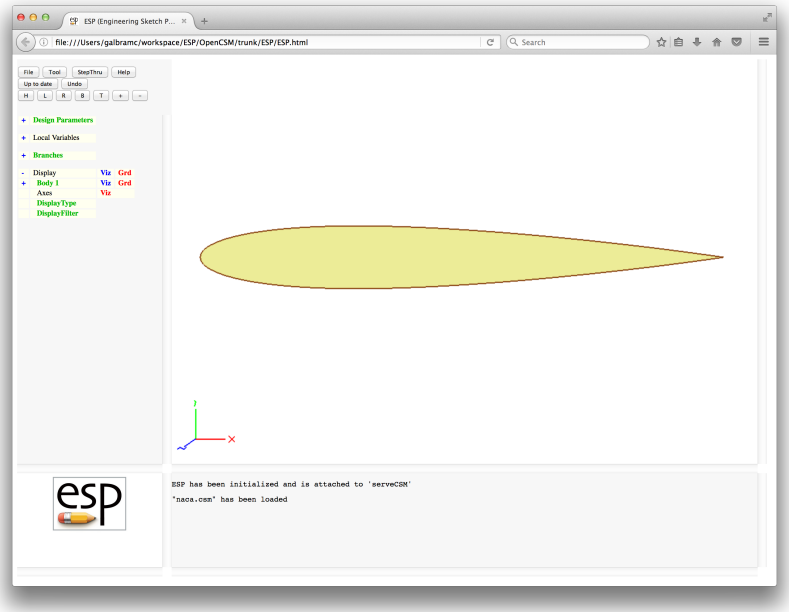
- Accessing/modifying analysis values
  - set/getAnalysisVal
- Analysis execution and outputs
  - pre/postAnalysis
  - getAnalysisOutVal
- DIRTY/CLEAN process
  - Tracking changes to inputs that impact outputs
- `capsGroup` attribute
  - Connecting geometry with analysis properties
- Suggested Exercises

NACA Airfoil Geometry

session3.5/naca.csm

```
# NACA design paramters
DESPMTR   thick     0.12      #frac of local chord
DESPMTR   camber    0.00      #frac of local chord

# Construct the airfoil
UDPRIM  naca   Thickness thick   Camber  camber
    ATTRIBUTE capsAIM $xfoilAIM;tsfoilAIM
```



Setting/accessing analysis values with pyCAPS

- Analysis values of an AIM are set or accessed with set/getAnalysisVal

session3.5/xfoil_1_AnalysisVal.py

```
# Load xfoil aim
print ("\n==> Loading xfoilAIM")
xfoil = myProblem.loadAIM(aim = "xfoilAIM",
                          analysisDir = "workDir_xfoil_1")

# Set Mach number
xfoil.setAnalysisVal("Mach", 0.5 )

# Print the modified mach number
mach = xfoil.getAnalysisVal("Mach")
print("\n==> Modified Mach =", mach)
```

Setting/accessing analysis values with pyCAPS

- Analysis values can be tuples/lists and toggled

session3.5/xfoil_1_AnalysisVal.py

```
# Print the default value of None
print("\n==> Default Alpha =", xfoil.getAnalysisVal("Alpha"))

# Set Alpha number
xfoil.setAnalysisVal("Alpha", 2.5 )
print("\n==> Modified Alpha =", xfoil.getAnalysisVal("Alpha"))

# Set list of Alpha
xfoil.setAnalysisVal("Alpha", [0.0, 3.0, 5.0, 7.0, 8.0] )
print("\n==> Modified Alpha =", xfoil.getAnalysisVal("Alpha"))

# Unset Alpha back to None
xfoil.setAnalysisVal("Alpha", None )
print("\n==> Unset Alpha =", xfoil.getAnalysisVal("Alpha"))
```

- Available analysis values in xfoil AIM documentation

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Running analysis with pyCAPS: Inputs

- Load AIM and set analysis values

session3.5/xfoil_2_Analysis.py

```
print ("\n==> Loading xfoilAIM")
xfoil = myProblem.loadAIM(aim = "xfoilAIM",
                          analysisDir = "workDir_xfoil_2")

print ("\n==> Setting analysis values")
# Set Mach and Reynolds number
xfoil.setAnalysisVal("Mach", 0.5 )
xfoil.setAnalysisVal("Re", 1.0E6 )

# Set list of Alpha
xfoil.setAnalysisVal("Alpha", [0.0, 3.0, 5.0, 7.0, 8.0] )
```

Running analysis with pyCAPS: preAnalysis

- Run preAnalysis to generate xfoil input files

  workDir_xfoil_2/caps.xfoil
  workDir_xfoil_2/xfoilInput.txt

session3.5/xfoil_2_Analysis.py

```
print ("\n==> Loading xfoilAIM")
xfoil = myProblem.loadAIM(aim = "xfoilAIM",
                          analysisDir = "workDir_xfoil_2")
```

```
# Run AIM pre-analysis
print ("\n==> Running preAnalysis")
xfoil.preAnalysis()
```

Running analysis with pyCAPS: Execution

- Execute xfoil in workDir_xfoil_2

session3.5/xfoil_2_Analysis.py

```
####### Run xfoil ####################
print ("\n\n==> Running xFoil......")

currentDirectory = os.getcwd() # Get current working directory
os.chdir(xfoil.analysisDir)    # Move into test directory

# Run xfoil via system call
os.system("xfoil < xfoilInput.txt > Info.out");

os.chdir(currentDirectory)     # Move back to top directory
#######################################
```

Running analysis with pyCAPS: postAnalysis

- Run postAnalysis to indicate completion and parse output files

session3.5/xfoil_2_Analysis.py

```
# Run AIM post-analysis
print ("\n==> Running postAnalysis")
xfoil.postAnalysis()
```

- Get results from the analysis with getAnalysisOutVal

```
# Retrieve Alpha, Cl and Cd
print ("\n==> Retrieve analysis results")
Alpha = xfoil.getAnalysisOutVal("Alpha")
Cl    = xfoil.getAnalysisOutVal("CL")
Cd    = xfoil.getAnalysisOutVal("CD")

print()
print("--> Alpha =", Alpha)
print("--> Cl    =", Cl)
print("--> Cd    =", Cd)
```

Multi analysis with the same AIM: Run

- Helper function for running the analysis

session3.5/xfoil_3_Analysis.py

```
def run_xfoil(xfoil):
    # Run AIM pre-analysis
    print ("\n==> Running preAnalysis")
    xfoil.preAnalysis()

    ####### Run xfoil ####################
    print ("\n\n==> Running xFoil......")

    currentDirectory = os.getcwd() # Get current working directory
    os.chdir(xfoil.analysisDir)    # Move into test directory

    # Run xfoil via system call
    os.system("xfoil < xfoilInput.txt > Info.out");

    os.chdir(currentDirectory)     # Move back to top directory
    ########################################

    # Run AIM post-analysis
    print ("\n==> Running postAnalysis")
    xfoil.postAnalysis()
```

Multi analysis with the same AIM: Alpha

- Compute polar for a range of angles of attack

session3.5/xfoil_3_Analysis.py

```
print ("\n==> Setting analysis values")
# Set Mach and Reynolds number
xfoil.setAnalysisVal("Mach", 0.5 )
xfoil.setAnalysisVal("Re", 1.0E6 )

# Set list of Alpha
xfoil.setAnalysisVal("Alpha", [0.0, 3.0, 5.0, 7.0, 8.0] )

# Run xfoil
run_xfoil(xfoil)

# Retrieve Alpha, Cl and Cd
print ("\n==> Retrieve analysis results")
Alpha = xfoil.getAnalysisOutVal("Alpha")
Cl    = xfoil.getAnalysisOutVal("CL")
Cd    = xfoil.getAnalysisOutVal("CD")

print()
print("--> Alpha =", Alpha)
print("--> Cl    =", Cl)
print("--> Cd    =", Cd)
```

Multi analysis with the same AIM: CL

- Switch to compute polar for a range of lift coefficients

session3.5/xfoil_3_Analysis.py

```
# Unset Alpha, otherwise it will be included in the next analysis
xfoil.setAnalysisVal("Alpha", None )

# Set specific Cl values instead
xfoil.setAnalysisVal("CL", [0.0, 0.1, 0.15, 0.3, 0.4] )

# Run xfoil
run_xfoil(xfoil)

# Retrieve Alpha, Cl and Cd
print ("\n==> Retrieve analysis results")
Alpha = xfoil.getAnalysisOutVal("Alpha")
Cl    = xfoil.getAnalysisOutVal("CL")
Cd    = xfoil.getAnalysisOutVal("CD")

print()
print("--> Alpha =", Alpha)
print("--> Cl    =", Cl)
print("--> Cd    =", Cd)
```

Analysis with geometric changes: Setup

- Setup analysis values

session3.5/xfoil_4_Camber.py

```
# Load geometry [.csm] file
filename = "naca.csm"
print ("\n==> Loading geometry from file \""+filename+"\"...")
naca = myProblem.loadCAPS(filename)

# Load xfoil aim
print ("\n==> Loading xfoilAIM")
xfoil = myProblem.loadAIM(aim = "xfoilAIM",
                          analysisDir = "workDir_xfoil_4")

print ("\n==> Setting analysis values")
# Set Mach and Reynolds number
xfoil.setAnalysisVal("Mach", 0.5 )
xfoil.setAnalysisVal("Re", 1.0E6 )

# Set list of Alpha
xfoil.setAnalysisVal("Alpha", [0.0, 1.0, 3.0] )
```

Analysis with geometric changes: Camber

- Execute sequence of cambers

session3.5/xfoil_4_Camber.py

```
# List of cambers to analyze
Cambers = [0.00, 0.01, 0.04, 0.07]

Alpha = []; Cl = []; Cd = []
for camber in Cambers:
    # Modify the camber
    naca.setGeometryVal("camber", camber)

    # Run xfoil
    run_xfoil(xfoil)

    # Append Alpha, Cl and Cd
    print ("\n==> Retrieve analysis results")
    Alpha.append(xfoil.getAnalysisOutVal("Alpha"))
    Cl.append(xfoil.getAnalysisOutVal("CL"))
    Cd.append(xfoil.getAnalysisOutVal("CD"))

print()
print("--> Cambers =", Cambers)
print("--> Alpha   =", Alpha)
print("--> Cl      =", Cl)
print("--> Cd      =", Cd)
```

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CAPS CLEAN/DIRTY Process

- Calling setGeometryVal marks geometry as DIRTY
  - Geometry always built just-in-time
- Calling setAnalysisVal or setGeometryVal marks the AIM as DIRTY
- CAPS does not execute analysis, cannot execute just-in-time
- Errors reported accessing getAnalysisOutVal if AIM is DIRTY
- Errors reported running preAnalysis if AIM is CLEAN

CAPS Successful Execution

- Execution without errors

session3.5/xfoil_5_CleanDirty.py

```
print("\n1. No Errors ", "-"*80)

# Set Mach and Reynolds number
xfoil.setAnalysisVal("Mach", 0.5 )
xfoil.setAnalysisVal("Re", 1.0E6 )

# Set list of Alpha
print("\n==> Setting alpha sequence")
xfoil.setAnalysisVal("Alpha", [0.0, 3.0, 5.0, 7.0, 8.0] )

# Run xfoil
run_xfoil(xfoil)

# Retrieve Cl
Cl = xfoil.getAnalysisOutVal("CL")
print("\n--> Cl    =", Cl)
```

CAPS DIRTY Analysis

- Trying call getAnalysisOutVal with DIRTY AIM due to setAnalysisVal

session3.5/xfoil_5_CleanDirty.py

```
print("\n2. DIRTY AnalysisVal Error ", "-"*80)

# Set a new alphas
print("\n==> Setting new alpha sequence")
xfoil.setAnalysisVal("Alpha", [1.0, 2.0] )

# Try to retrieve Cl without executing pre/postAnalysis
print("\n==> Attempting to get Cl")
try:
    Cl = xfoil.getAnalysisOutVal("CL")
    print("\n--> Cl    =", Cl)
except CAPSError as e:
    print("\n==> CAPSError =", e)
```

CAPS DIRTY Geometry

- Trying call getAnalysisOutVal with DIRTY AIM due to setGeometryVal

session3.5/xfoil_5_CleanDirty.py

```
print("\n3. DIRTY GeometryVal Error ", "-"*80)

# Modify a geometric parameter
print("\n==> Modifying camber")
myProblem.geometry.setGeometryVal("camber", 0.07)

# Try to retrieve Cl without executing pre/postAnalysis
print("\n==> Attempting to get Cl")
try:
    Cl = xfoil.getAnalysisOutVal("CL")
    print("\n--> Cl    =", Cl)
except CAPSError as e:
    print("\n==> CAPSError =", e)
```

CAPS DIRTY postAnalysis

- Trying to call getAnalysisOutVal without calling postAnalysis

session3.5/xfoil_5_CleanDirty.py

```
print("\n4. DIRTY pre- but no postAnalysis Error ", "-"*80)

# Modify mach number
print("\n==> Modifying Mach")
xfoil.setAnalysisVal("Mach", 0.3 )

# Run AIM pre-analysis
print ("\n==> Running preAnalysis but not running postAnalysis")
xfoil.preAnalysis()

# Retrieve Cl
print("\n==> Attempting to get Cl")
try:
    Cl = xfoil.getAnalysisOutVal("CL")
    print("\n--> Cl    =", Cl)
except CAPSError as e:
    print("\n==> CAPSError =", e)
```

CAPS CLEAN Error

- Calling preAnalysis with a CLEAN AIM

session3.5/xfoil_5_CleanDirty.py

```
print("\n5. CLEAN Error ", "-"*80)

# Don't modify any analysis or geometry values

try:
    # Run xfoil
    run_xfoil(xfoil)
except CAPSError as e:
    print("\n==> CAPSError =", e)
```

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The capsGroup attribute – Masstran Wing

capsGroup attribute

- Tags groups of BODY/FACE/EDGE/NODE
  - Entities with same capsGroup value are in the group
- Specific use of capsGroup is AIM dependent

session3.5/masstran_6_f118_Wing.py

```
filename = "f118-C.csm"
print ("\n==> Loading geometry from file \""+filename+"\"...")
myProblem.loadCAPS(filename, verbosity=0)

# Load the masstran aim with the wing
masstran = myProblem.loadAIM(aim = "masstranAIM",
                             analysisDir = "workDir_masstranWing",
                             capsIntent="wing")
```

The capsGroup attribute – Masstran Wing

- Wing FACEs tagged with "wing:faces"
- masstranAIM material and properties for "wing:faces"

session3.5/f118-C.csm

```
# Wing
SET       wing:span      sqrt(wing:aspect*wing:area)
SET       wing:chord     wing:area/wing:span

BOX  wing:xroot  -wing:span/2  wing:zroot   wing:chord  wing:span  wing:chord*wing:thick
    ATTRIBUTE capsGroup  $wing:faces
```

session3.5/masstran_6_f118_Wing.py

```
# Define material properties
unobtainium = {"density" : 7850}

# Set the material
masstran.setAnalysisVal("Material", ("Unobtainium", unobtainium))

# Define shell property
shell = {"propertyType"      : "Shell",
         "material"          : "Unobtainium",
         "membraneThickness" : 0.2}

# Associate the shell property with capsGroups defined on the geometry
masstran.setAnalysisVal("Property", ("wing:faces", shell))
```

The capsGroup attribute – Masstran F-118

- capsGroups on all FACEs

session3.5/f118-C.csm

```
BOX  wing:xroot  -wing:span/2  wing:zroot   wing:chord  wing:span  wing:chord*wing:thick
    ATTRIBUTE capsGroup  $wing:faces
```

```
BOX  htail:xroot  -htail:span/2  htail:zroot   htail:chord  htail:span  htail:chord*htail:thick
    ATTRIBUTE capsGroup  $htail:faces
```

```
BOX  vtail:xroot  0  vtail:zroot   vtail:chord  vtail:chord*vtail:thick  vtail:span
    ATTRIBUTE capsGroup  $vtail:faces
```

```
BOX  0  -fuse:width/2  -fuse:height/2   fuse:length  fuse:width  fuse:height
SELECT face 1
    ATTRIBUTE capsGroup  $fuse:nose
SELECT face 2
    ATTRIBUTE capsGroup  $fuse:tail
SELECT face 3
    ATTRIBUTE capsGroup  $fuse:side
SELECT face 4
    ATTRIBUTE capsGroup  $fuse:side
SELECT face 5
    ATTRIBUTE capsGroup  $fuse:side
SELECT face 6
    ATTRIBUTE capsGroup  $fuse:side
```

The capsGroup attribute – Masstran F-118

- Properties assigned to capsGroups

session3.5/masstran_7_f118.py

```
# Define material properties
unobtainium = {"density" : 7850}
madeupium   = {"density" : 6890}

# Set the materials
masstran.setAnalysisVal("Material", [("Unobtainium", unobtainium),
                                     ("Madeupium", madeupium)])

# Define shell properties
shell_1 = {"propertyType"      : "Shell",
           "material"          : "unobtainium",
           "membraneThickness" : 0.2}

shell_2 = {"propertyType"      : "Shell",
           "material"          : "madeupium",
           "membraneThickness" : 0.3}

# Associate the shell property with capsGroups defined on the geometry
masstran.setAnalysisVal("Property", [("wing:faces" , shell_1), ("htail:faces", shell_1),
                                     ("fuse:nose"  , shell_1), ("fuse:tail"  , shell_1),
                                     ("vtail:faces", shell_2), ("fuse:side"  , shell_2)])
```

The capsGroup attribute – AVL Plane Vanilla

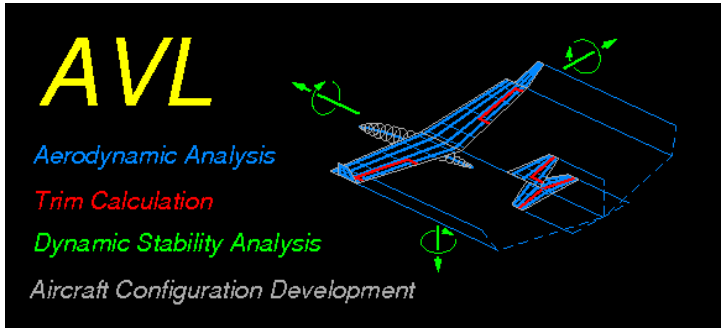
- AVL - Vortex Lattice Method: Geometry defined by airfoils
- capsGroup groups airfoils into surfaces

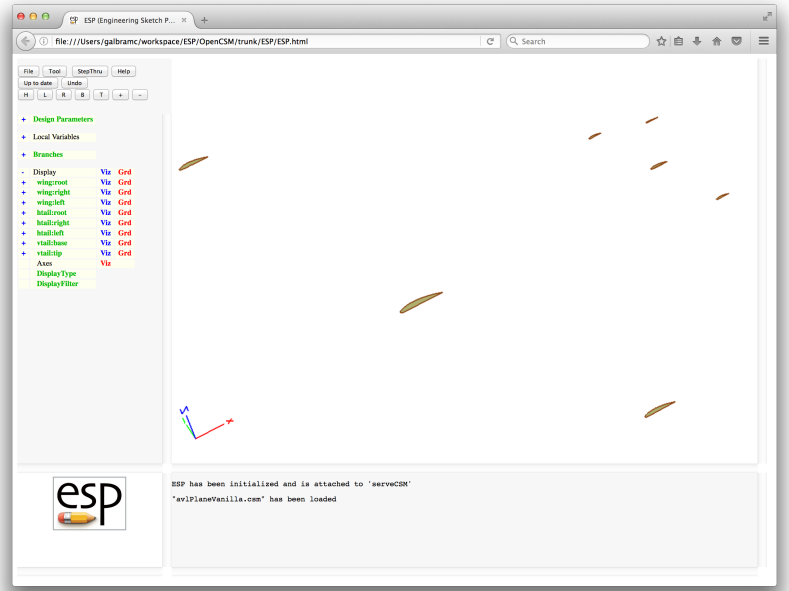
session3.5/avlPlaneVanilla.csm

```
UDPRIM    naca  Thickness wing:thick  Camber wing:camber
SCALE     wing:croot
ROTATEX   90         0          0
TRANSLATE wing:xroot     0     wing:zroot
    ATTRIBUTE capsGroup          $Wing
```

```
UDPRIM    naca  Thickness wing:thick  Camber wing:camber
SCALE     htail:croot
ROTATEX   90         0          0
TRANSLATE htail:xroot     0     htail:zroot
    ATTRIBUTE capsGroup $Htail
```

```
UDPRIM    naca  Thickness vtail:thick
SCALE     vtail:croot
TRANSLATE vtail:xroot     0     vtail:zroot
    ATTRIBUTE capsGroup $Vtail
```





The capsGroup attribute – AVL Plane Vanilla

- VLM meshing parameters defined via capsGroups

session3.5/avl_8_PlaneVanilla.py

```
# Load avl aim
print ("\n==> Loading avlAIM")
avl = myProblem.loadAIM(aim = "avlAIM",
                        analysisDir = "workDir_avl")

print ("\n==> Setting analysis values")
avl.setAnalysisVal("Alpha", 1.0)

# Set meshing parameters for each surface
wing = {"numChord"     : 4,
        "numSpanTotal" : 24}

htail = {"numChord"     : 4,
         "numSpanTotal" : 16}

vtail = {"numChord"     : 4,
         "numSpanTotal" : 10}

# Associate the surface parameters with capsGroups defined on the geometry
avl.setAnalysisVal("AVL_Surface", [("Wing" , wing ),
                                   ("Htail", htail),
                                   ("Vtail", vtail)])
```

Suggested Exercises

Thickness

- Generate airfoil polars for a range of airfoil thicknesses
  - Start from a copy of session3.5/xfoil_4_Camber.py

- Create your own